

WHAT IS CLAIMED IS:

1. A method for assembling a combustor for a gas turbine engine, the combustor including a swirler assembly, said method comprising:

machining material to form a domeplate;

positioning a sealplate including an overhanging portion against the domeplate;

securing the sealplate in position relative to the domeplate with a welding process; and

welding the swirler assembly to the domeplate.

2. A method in accordance with Claim 1 wherein machining material to form a domeplate further comprises forming an opening extending through the domeplate such that the opening is defined along a first side of the domeplate by a chamfered edge.

3. A method in accordance with Claim 2 wherein forming an opening extending through the domeplate further comprises counter-boring an edge defining the opening along an opposite second side of the domeplate.

4. A method in accordance with Claim 2 further comprising coupling a swirler to the combustor such that at least a portion of the swirler extends through the domeplate opening.

5. A method in accordance with Claim 1 wherein securing the sealplate in position relative to the domeplate further comprises securing the sealplate in position to facilitate aligning the swirler relative to the domeplate.

6. A method in accordance with Claim 1 wherein securing the sealplate in position relative to the domeplate further comprises securing the sealplate in position such that a gap is defined between the sealplate overhang portion and the domeplate.

7. A method in accordance with Claim 1 further comprising coupling a baffle to the sealplate using a brazing process.

8. A combustor for a gas turbine engine, said combustor comprising:

a swirler assembly; and

a dome assembly comprising a sealplate and a domeplate, said sealplate welded to said domeplate and comprising an overhang portion and an integrally-formed body, said sealplate welded to said domeplate such that a gap is defined between said domeplate and said sealplate overhang portion, said swirler assembly welded to said domeplate.

9. A combustor in accordance with Claim 8 wherein said domeplate comprises an upstream side, a downstream side, and an opening extending therebetween, at least one of said upstream and downstream sides comprises a chamfered edge that defines said opening.

10. A combustor in accordance with Claim 8 wherein said domeplate comprises an upstream side, a downstream side, and an opening extending therebetween, at least one of said domeplate upstream and downstream sides comprises a counter-bored edge that defines said opening.

11. A combustor in accordance with Claim 10 wherein at least a portion of said sealplate is secured within said counter-bored edge, said counter-bored edge facilitates aligning said swirler assembly relative to said domeplate.

12. A combustor in accordance with Claim 8 further comprising a baffle brazed to said sealplate.

13. A combustor in accordance with Claim 8 wherein said swirler assembly comprises at least a secondary swirler welded to said sealplate and a primary swirler coupled to said secondary swirler such that said primary swirler is free to move against said secondary swirler..

14. A gas turbine engine comprising a combustor comprising a dome assembly, at least one injector, and an air swirler, said dome assembly comprising a sealplate and a domeplate, said sealplate welded to said domeplate and comprising a body and an overhang portion extending integrally from said body, said sealplate welded to said domeplate such that a gap is defined between said domeplate and said sealplate overhang portion, said swirler assembly welded to said domeplate, said at least one injector coupled to said dome assembly.

15. A gas turbine engine in accordance with Claim 14 wherein said domeplate comprises an upstream side, a downstream side, and an opening extending therebetween, said opening sized to receive at least a portion of said air swirler therethrough, at least one of said domeplate upstream and downstream sides comprises a chamfered edge that circumscribes said opening such that said edge defines said opening.

16. A gas turbine engine in accordance with Claim 14 wherein said domeplate comprises an upstream side, a downstream side, and an opening extending therebetween, said opening sized to receive at least a portion of said air swirler therethrough, at least one of said domeplate upstream and downstream sides comprises a counter-bored edge that circumscribes said opening such that said edge defines said opening.

17. A gas turbine engine in accordance with Claim 15 wherein said counter-bored edge is sized to receive at least a portion of said sealplate therein such that said counter-bored edge facilitates aligning said swirler assembly relative to said domeplate.

18. A gas turbine engine in accordance with Claim 14 wherein said combustor further comprises a baffle welded to said sealplate and extending downstream from said domeplate.

19. A gas turbine engine in accordance with Claim 14 wherein said swirler assembly comprises at least a secondary swirler welded to said sealplate and a primary swirler coupled to said secondary swirler.